

Suppose we want a satellite to revolve around Earth 7.5 times a day. What should the radius of its orbit be?

$$\text{For Earth } R^3/T^2 = 10 \times 10^{12} \text{ m}^3/\text{s}^2 = \frac{G M_E}{4\pi^2}$$

$$\text{Now we want } T = 24 \text{ h} / 7.5 \text{ times} = 11520 \text{ s}$$

$$R^3 = (10 \times 10^{12}) \cdot (11520)^2$$

$$R = \sqrt[3]{10 \times 10^{12} (11520)^2}$$

$$= 1.09 \times 10^7 \text{ m}$$

Find the orbital speed of a satellite orbiting about a planet, if the mass of the planet is  $5.67 \times 10^{26}$  kg and the radius of the orbit is 250,000 km.

Note:  $G = 6.67 \times 10^{-11} \text{ N}\cdot\text{m}^2/\text{kg}^2$

$$\frac{mv^2}{r} = F_g = G \frac{mM}{r^2}$$

Centripetal Force =  $m_{\text{Sat}} \cdot v^2/R_{\text{Orbit}} = \text{Gravitational Force} = G \cdot M_{\text{Planet}} \cdot m_{\text{Sat}}/R^2_{\text{(orbit)}}$

$$v^2 = \frac{G \cdot M_{\text{PLANET}}}{R}$$

$$v = \sqrt{\frac{G \cdot M_{\text{PLANET}}}{R}}$$

$$= \sqrt{\frac{(6.67 \times 10^{-11}) \cdot 5.67 \times 10^{26}}{250000 \times 10^3 \text{ m}}}$$

$$= 1.22 \times 10^3 \frac{\text{m}}{\text{s}}$$

$$M_{\text{Mars}} = 6.42 \times 10^{23} \text{ kg}, \quad R_{\text{Mars}} = 3.37 \times 10^6 \text{ m}$$

$v_{\text{esc Mars}} = ?$

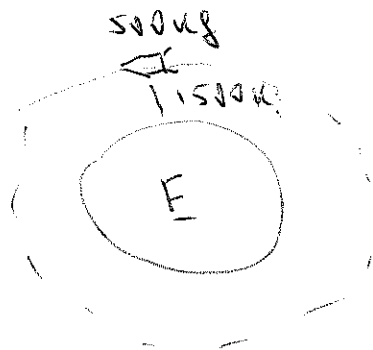
**68. ORGANIZE AND PLAN** The escape speed from the surface of Mars can be found using Eq. 9.6, replacing the Earth's mass and radius with the mass and radius of Mars.

*Known:*  $M_{\text{Mars}} = 6.42 \times 10^{23} \text{ kg}$ ,  $R_{\text{Mars}} = 3.37 \times 10^6 \text{ m}$ .

**SOLVE** The escape speed from the surface of Mars is [Eq. 1]

$$v_{\text{esc}} = \sqrt{\frac{2GM_{\text{Mars}}}{R_{\text{Mars}}}} = \sqrt{\frac{2(6.67 \times 10^{-11} \text{ N} \cdot \text{m}^2/\text{kg}^2)(6.42 \times 10^{23} \text{ kg})}{3.37 \times 10^6 \text{ m}}} = \underline{5.04 \text{ km/s}}$$

**REFLECT** This is less than half the escape speed from the Earth's surface, which is 11.2 km/s.



**61. ORGANIZE AND PLAN** Use Eq. 9.7 to calculate the total energy of the satellite. The radius of the satellite's orbit is [Eq. 1]  $r = R_E + h$ , where  $h$  is the height of the satellite above the surface of the Earth.

*Known:*  $m = 500 \text{ kg}$ ;  $h = 1500 \text{ km}$ ;  $M_E = 5.97 \times 10^{24} \text{ kg}$ ;  $R_E = 6.37 \times 10^6 \text{ m}$  (Appendix E)

**SOLVE** Using Eq. (1) in Eq. 9.7 gives [Eq. 2]

$$E = -\frac{GM_E m}{2(R_E + h)} = -\frac{(6.67 \times 10^{-11} \text{ N} \cdot \text{m}^2/\text{kg}^2)(5.97 \times 10^{24} \text{ kg})(500 \text{ kg})}{2(6.37 \times 10^6 \text{ m} + 1.5 \times 10^6 \text{ m})} = -1.26 \times 10^{10} \text{ J}$$

**REFLECT** Notice that the height was converted to SI units to be consistent with the other quantities in the calculation.